

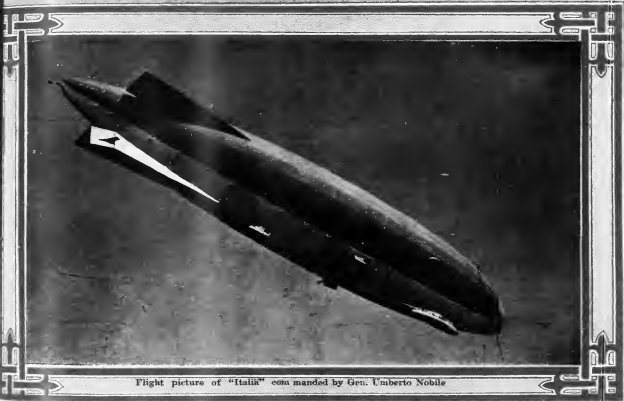
AVIATION

The Oldest American Aeronautical Magazine

MAY 28, 1928

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Flight picture of "Italia" commanded by Gen. Umberto Nobile

VOLUME
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22

Special Features

The Stinson "Junior"
The Detroit Air-Olympics
The Szekely SR-3 Air Cooled Engine

AVIATION PUBLISHING CORPORATION
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How is production of EAGLEROCKS

Colorado Springs factory, buildings are nearing completion. In the meantime temporary production quarters are being used with results, and more Eaglerocks are being turned out than the two a day output of the old Denver factory. We are rapidly catching up with our back orders.

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In about sixty days the new factory will be close to a year in hand production schedule which should be sufficient to run for the ever growing demand for Eaglerock for many months.

Significantly with doubt this motor must be present regardless of the fact that it is one of all our engines, which up to 1935 is

PRISON
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Eaglerocks are now being built and tested at the foot of Pikes Peak which means still increased altitude and unusual altitude performance.



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THANK YOU for watching AVIATION

May 26, 1935

CAMINEZ ... CLIMB

Airplane (1) is powered by a Fairchild Caminez Engine, taking off at the same time as airplane (2), which is powered by a conventional type of engine of the same horsepower. In one minute after the take-off, the Caminez-powered plane (1) will have climbed 40% higher (at 125% greater angle of climb) than plane (2). Why?

A CAMINEZ POWERED plane will climb 40% higher, in a given time, with so much less forward speed that the angle of climb will be 125% greater than that of a plane powered by the most type of engine of the same horsepower. For this reason, a Caminez-powered plane will turn to climb after less "grossed-out" and can consequently get out of skirts where a plane powered by the conventional type of engine could never take-off. The unusual ability of the Caminez Engine to climb is the result of the high efficiency obtained by the half speed propeller. This half-speed propeller efficiency is made possible by the use of the patented cam drive and without the disadvantages of reduction gears.

Henceforth, low weight per horse power has been developed in aviation engines by increased propeller speed through greater engine r.p.m. But, experience has proven that a high-speed propeller does not give high maximum efficiency — particularly during climb and take-off, when high thrust power is so essential.

Consequently, to obtain maximum thrust and at the same time overcome the lack

in efficiency of the high-speed propeller for conventional aircraft — it is necessary either to use propeller reduction gears with the ordinary type of engine, or to use the Fairchild Caminez Engine.

"Climb" is but one of the outstanding features of the Fairchild Caminez. The complete story of this remarkable engine is too big for a single advertisement. Write for the Caminez Engine Book. The Fairchild Caminez Engine Corporation, Farmingdale, Long Island, New York.



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CAMINEZ ENGINES



THANK YOU for watching AVIATION

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THE STARLING

AN ACHIEVEMENT AND AN INNOVATION IN DESIGN

Inaugurating new principles in design. Although technique is very important in the flying qualities of the airplane, the fundamental principles of design have not been departed from, but those things have been incorporated that have been neglected in the present day plane.

TO attain such airworthiness requires something more than fine material and precise workmanship. That something more is clean cut design with safety and pay load the practical objectives.

Safety is assured by the remarkable balance of the plane. Of equal importance is the fact that the STARLING has perfect control even below the stalling speed and all controls operate with the least possible physical effort. Another important feature of the STARLING is the landing gear, which withstands any unusual strains and is practically noiseless in its operation.

Next is the combination of airfoils that gives the plane a remarkable degree of stability that is found in no other plane. The STARLING embodies all of the flying characteristics of the high powered scout, using an economical power unit of only 90 h.p. This is made possible by the

correctness of design and its construction allows it to perform every maneuver known to the scout type. It can be flown and landed equally well from either cockpit. A student properly instructed on the STARLING is able to safely pilot most types of planes, without the necessity of instruction on the intermediate type of advanced training machine.

Still the STARLING maintains a landing speed of only 30 miles per hour and a maximum speed of 104 m. p. h.

This remarkable plane was designed by an expert designing engineer, who has at this time two other planes to his credit that are being built under quantity production with an approved type certificate from the Department of Commerce.

The eight years of actual experience in the aircraft industry is the basis on which the STARLING was produced from the designs of Orville H. Hickman.



Starling Aircraft COMPANY

Minneapolis, Minnesota

CLARK YOG for marketing ATTENTION

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"A good engine"—

alone it carries on for hours without musing, for miles without stopping. "A good engine"—seemingly without fatigue, a power-plant that knows no exhaustion, whose performance is so consistent that it seems to have no limits.

"A good engine"—but back of that popular comment is a story of conception, design, workmanship, assembly, tests by the hundred, and infinite care to every detail that makes a good engine, not by chance, but by deliberate attention.

Wright Engines were born in the Great War and reared in the atmosphere of successful air activities. Their tests came every day. They proved their ability in the vital ebb and flow

speed, maneuverability and endurance. The Wright organization placed unsurpassed methods of design and workmanship at the disposal of foreign governments and the United States and developed aircraft engines which were unrivaled in quality of workmanship and performance.

With this background, followed by further development for nearly 14 years, it is but logical that these engines perform as they do. Believing that the entire aeronautical industry will be interested in knowing how carefully and surely Wright Engines are built, special processes will be described in a series of advertisements of which this is the first.

of war necessity. Their performance helped make "aces" of pilots whose names are still remembered.

Wright was drafted in the early war-time days to produce an engine capable of meeting war-time necessities for durability, sustained



The Oldest American Aeronautical Magazine

Vol. XLIV

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No. 22

Give Them Service

I PLANS and preparations now under way are carried out according to schedule, there should be at least 500 standard airports in operation by the end of the year. Just what the equipment provided by these airports will be is a matter for speculation, but it is simply hoped that each will at least provide suitable facilities for the servicing of airplanes.

The problem of service is one which is becoming more and more important as each new plane takes the air. It is the time of Lindbergh's great flight that were the enough planes in operation to cause the maintenance to consider that phase. But now that production is being doubled and tripled it is different. The plane owner not only wants service but demands it and the manufacturer who does not immediately take steps to render service to his customers will be hit by his own lowering.

The rendering of service is one engine has proved to be one of the best sales arguments in the automobile industry. There is no reason to believe it is not so in the aeronautical industry. Heretofore the plane manufacturer has had only a product to sell but now he has service to sell also. He has got to consider the question of establishing authorized service stations throughout the country if he intends to keep his customers, and incidentally get new ones.

A satisfied customer is not only a prospect for next year's model, but he is also a splendid advertising medium. However, if he is not able to obtain service he will not be satisfied for very long and the subsequent loss to the manufacturer will be double.

Rescuing the Bremen

ONE OF the outstanding features of the West to West coasting of the North Atlantic that has astonished it from all other successful crossings is the total disregard for the weather of the plane by those who go and it is the greatest sailing plane. When Lindbergh landed in Paris his first season, was for his plane and as soon as he had obtained sufficient rest from his trial he went to Le Bourget Field and saved for his plane. And when Chamberlin damaged the "Columbia" on the second landing in Germany he stayed right with his plane until it was repaired and ready for additional flights.

However with the case of the "Bremen" it was a different story. When it was found impossible to remove

it from Gretna Island the crew left it there and came out to civilization in another plane. That was a most serious matter in view of the fact that the world in general desired to be before its prize upon the gallant crew for their great accomplishment. But instead of going back for their plane at the earliest opportunity they started off on a country wide tour and left the task of removing the Bremen from Gretna Island to the U. S. Government.

Just how the Government's aid was solicited is not quite clear and it does not matter. But it is rather surprising that the head of the U. S. Army Air Corps should take charge of what is nothing less than a rescue operation. Of course it is realized that the Chief of the Army Air Corps is the type of man who does not spend all his time in his office, and goes out and really flies. And it is also realized that he would be the first to render aid to another aviator. However his position is one of great responsibility and his participation in such a venture does not seem necessary. Undoubtedly he is carrying out the orders of his superior, and if so then they are to be criticized for their line of action. To lead assistance in getting the Bremen out of Gretna Island is a worthy undertaking for the U. S. Army Air Corps, yet at the same time it is a bit surprising that the Army's crew should carry on leaving the country and leave it to the officer of rank in the Air Corps to head the rescue expedition.

The Twentieth Anniversary

THE EIGHTH day of next month will be one of celebration in English aeronautical circles for it marks the twentieth anniversary of the first airplane flight over British soil. The pilot on that memorable aerial venture was A. V. Roe and the plane was a twin-engine biplane of his own construction fitted with a 24 hp. Anzani engine. Like the late Wilbur Wright's famous flight at Kitty Hawk, Mr. Roe's flight at Brooklands was not long in duration as he only covered a distance of sixty yards at the height of but a few feet. Nevertheless it was the beginning of successful heavier-than-air development in England.

Mr. Roe is still actively engaged in that work of development and his Avro planes are well known throughout the entire aeronautical world. Aviators take this opportunity to not only congratulate Mr. Roe for his splendid accomplishment twenty years ago but also for his equally splendid accomplishments since that historic date.



Wright Cyclone
Aeronautical
Engine

WRIGHT

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TELEGRAMS FOR ordering AVIATION



Wright Whirlwind
Aeronautical
Engine

*That's why
More Pilots fly them!*

The Szekely SR-3 Air Cooled Engine

New Three Cylinder Radial is Rated at
40 H.P. at 1800 R.P.M.

THE O. E. Sweeney Corp. of Holland, Mich., has been consulting and industrial engineers to various manufacturers for a number of years and as a result of the experience gained in this work, it reached the conclusion about six months ago that there would be an ever increasing sale for simple, light weight, reliable, air cooled aircraft engine-welding units of the exclusive features which have worked out so well on equipment it has designed for others.

Apparently it is intended, the design of a three cylinder radial, air cooled engine for light aircraft. To be followed in the near future by a five cylinder, wing mounted, all of the parts of the three cylinder engine.

The extremely simple outward appearance of the three-spline design immediately commands it to approval, for the only visible moving parts are the push rods and roller arms, all remaining parts being entirely motionless. The engine has a 4½ in bore by 4½ in stroke, giving a total piston displacement of 100 cc in. Operating at the rated speed of 1800 r.p.m., 100 cc is converted, as a matter of fact, actual brake



1945 and 1946 show that the output develops over time. The cost-output ratio is 4.6 to 1.

The mounting flange for the engine consists of a round or shaped plate fitted rear the engine bracket on the plane. The engine is held in place by six tie rods, each of which pass axially through the engine, so that the front cover, main case, rear cover, and mounting bracket are assembly bolted together.

The cylinders are aluminum machined gray iron, with cylinder head face integral and are held to the crankcase by 16 bolts. Each cylinder has a skirt or piston guide extension approximately 25% into the crankcase. Intake and exhaust valves are located in the cylinder head. The cylinder is provided with a compression ring flange to provide an airtight seal between the ring and the cylinder wall. Valve stems are part of the crankcase and are at 45 deg. angle. Two valves are used in each cylinder, one intake, one exhaust. Each valve stem, machined to 15 mm diameter, is ground to 15 mm diameter for 15 mm of its length. The valves are 1.125 in. in diameter by 31/32 in. high. The valves are actuated from the cam through follow-up pins and leaf-spring rods and become hollow drop forged rods. The valves are ground to 15 mm diameter for 15 mm of their length. After adjustment, for a run screw.

The lubricating system is interesting. A diesel-type gear pump is driven by a bronze output gear directly from the crankshaft. Four gear pumps are used in the pump pack. The upper set removes oil through a 3% oil line from the crank, in the phase and delivers it under pressure to the main bearing, the piston and the cylinder walls. The main bearing pump and cylinder wall pump. The other pump is a scavenging pump and also slightly larger than those of the other pump. This picks up the oil from the lower part of the crank to which scumline runs, and delivers it through a 3% oil line to the oil tank. An interesting innovation from last year is that the oil of the relief valve and the take off for the oil pump are placed at the rear of the engine so that oil pressure remains in the rear after the engine is stopped.

The pressure at the pump is approximately 40 psi, and the scumline goes at the rear of the engine. This is an



Left: (a) — only parts, drop forged, heat treated crankshaft with SKF roller bearings. Center, the master connecting rod pin (b); (c) drop rod assembly. Right, showing the cone operation of the SN-3. Each cone is driven by an independent gear and controls two pistons.

• **multigrade refigurant**, as it shows the fact that each part of the engine is lubricating different lubrication as long as oil remains above the groups. A group pressure of 4 lb per sq in. is 18 lb per sq in. is recommended, which means the pump pressure usually noted on other engines is 48 lb per sq in. of 18 per sq in.

[illegible]

The crankshaft shaft is 3160 S-A E material, having a diameter of 1 37/64 in. The connecting rod bearing on the outside is 1 7/16 in. by 2 in. The crankshaft is a single throw design, having, in each counterweight, are added to assist in balancing at the crank shaft. The piston-rod thrust

bearing is an I.F.F. bearing of the Deep groove, radial ball bearing type and is taken directly between the front of the crankshaft and the front gear cover of the engine itself. The propeller hub is steel and heat treated for maximum strength. Each propeller hub is lapped to the crankshaft of its own engine, in addition to the precision operation of grinding both the hub and the crankshaft.

Connecting rod members are drop-forgings heat treated for toughness and machined to lower. The upper end of the connecting rod is bronze bushed. The lower bearing, consisting of a cylinder cast on an angle with the axis of the bearing and contact with a bronze bearing at the external portion of which is bushed lead and free to revolve around the crank pin. Oil pressure is delivered to the main connecting rod bearing and is seen in bushed lead contact with the rod bearing. This assembly is secured securely in cast in place bearing. The presence of complete lubrication and larger bearing surface is proportionate to the load placed upon these two mechanical portions of a motor and with most pistons rods separated from it.

The pistons are of aluminum alloy, solid skirt type, each equipped with three 1/8 in. piston rings. The piston pin is hollow, hardened and ground and fastened to the aluminum center at either end, but of the full floating type, in that it

Continued on page 2531



Left, one of the two aluminum alloy mainframes with cam roller gear in place. Center, the cylinder assembly. Note the oil pump drive gear. Right, the oil pump and assembly. It will be driven by the cam roller gear directly from the crankshaft.

The Stinson "Junior"

SM-2 is Almost Identical to the Detroitier Except that it is Smaller and is Powered with a 110 Hp. Warner Engine

ONE OF the first planes to get into production with the new Warner "Search" engine is the Stinson "Junior" or SM-2 cabin monoplane manufactured by the Stinson Aircraft Corp., Northridge, Mich. The plane is almost identical to the standard Stinson Detroitier, except that it is somewhat smaller, being powered with a 110 hp. engine instead of the 220 hp. Wasp. The closed cabin seats four people under the wing, which turn in with the top of the fuselage, increasing the head room. The wing is of semi-cantilever design supported by lift struts from the bottom of the fuselage. A new feature in the Stinson Junior, not found in other Stinson designs, is the wheel tail shield mounted on an oleo shock absorber leg. The first plane of this type was completed in three weeks, while the design work was completed in three weeks previous to that. William C. Napier, chief engineer of the Stinson Aircraft Corp., is the designer of the plane, which is intended to be a quality product for the sportsman, business man, or private owner.

The cabin of the SM-2 is quite wide, with four wheel chairs, properly upholstered, two in front and two behind. The necessary baggage can be stowed under these chairs. The inside of the cabin is trimmed with grove leather to match the upholstery on the chairs. Two doors, one on each side of the fuselage and extending the full depth of the cabin, are provided for entrance or exit. Steps in the leading gear struts make these doors easily accessible. The windows of the cabin are fitted with shatter-proof glass. The front window can be raised or lowered by means of a bell crank, thus allowing the pilot good vision in freezing weather. The engine and main cabin are on such a position that the pilot can look directly over the engine with the tail wheel as the ground. A skylight of Pyralis, in the center section of the wing, gives proper vision ahead and to the rear.

The plane is fitted with dual side by side stick control, with two control sticks and two sets of rubber pedals. The ailerons are worked by a combination of bell cranks, cables, and push pull rods. Flaps actuate the rubber, while the elevator and stabilizer are controlled by push pull tubes.

Three tubes are supported in special graphite bearings and require no lubrication for a period of about two years. In this way the control bearings may be forgotten and still be in the fuselage as required. The stabilizer cable consists of a lever and rubber arrangement. The ailerons can be made simply and quickly by one section from a plate. An adjustment of 2 1/2 deg. is provided in wheel position. From wheel, directly raising for any amount of landing position. This control, incidentally, is located in the middle of the cabin where it can be reached by a pilot or student. Although the plane was not designed as such it should prove a good training plane, for there are but a tendency towards closed while the training plane based on the experience that the student will learn by a closed plane and will therefore be better off starting in a model. The Stinson Junior lands at 60 m.p.h. with its load, which should prove slow enough for the landing of pilots with a minimum of risk.

Fuselage Structure of Seamless Steel

The fuselage structure is made up of seamless steel tubing. It is rectangular in section with the tubes of the side rails welded into a modified Warren truss. The tubes in the top and bottom provide frame a modified Pratt beam. The air used is 1000-D throughout, with the exception of center section tubes and vertical members around the door struts. The diagonal tubes, above and below the cabin, connect the front wing spar, almost at their midpoints, to the one tube in connection with the other tube welded to it. It is not cut through both tubes with a flat plate inserted in between. All the tubes are welded to this plate, reinforcing it. The stress caused by the lift struts attached to the lower fuselage is carried across the fuselage by 1 1/2 in. strips of steel used in tension between the main struts. One of these flat strips is on each side of a tube that forms part of the fuselage truss. The strips are welded at the end with a hole at the point through which the compression

from three strips than the part of the struts for the support of each set of lift struts. The rear turtle deck of the fuselage is back of a spar firing mounted on steel tube frame, giving a well rounded and correct section at all points.

The wing is of wood construction with metal fittings and steel struts. The two spars are of solid spruce rived in 1 in. section. The spacing is the same on both forward and rear spars, the spars varying only in depth, which can be increased production costs. There is no strutting at all struts, three about at the same point on both forward and rear members. A Clark Y section is used having rib of



Side view of the new Stinson "Junior" in flying position.

wood construction with square spruce cap strips and cross members joined with plywood gussets nailed and glued to the sides. The compression members between the spars consist of steel tubes welded into the faces of a Warren truss. These are constructed of two horizontal compression members with the truss member welded between them. The truss members consist of a single tube bent back and forth between horizontal members to form the diagonal bracing. These are welded to the horizontal members at each end. Fillets are welded to the ends of the compression tubes to take the loads square forward for the drag bracing. As there is no compression tube at the top and one at the bottom, bracing each spar, this system of bracing works out very nicely, allowing very simple fittings for the attachment of the diagonal bracing. The ailerons are mounted on a combination of three compression struts curved behind the rear spar then they form the aileron bracing. In front of the aileron leading the wing is finished with a false spar made up of a steel channel channel fastened in place by wood screws. The leading edge of the wing is covered with sheet duralumin in top and bottom in front of the forward spar. The leading edge is a strip of duralumin fitted along its length between the ribs fitted into the corner of the field. The wing tip consists of a large diameter tube bent to conform with the planform of the wing. To provide for the turning of the tip, small diameter tubes are welded to this tube. These small tubes are approximately one quarter inch in diameter and, through providing some very delicate welding work, give a light and strong wing tip.

The wings are supported by two lift struts on each side of the fuselage, one to each spar. The struts are of steel tubing fitted with ball joints to give an R-100 section. The forward struts are adjustable at the lower end to provide for raising the wing. At a point near the bottom of the forward strut, the leading gear is attached. This point is braced by an additional member to the lower fuselage member in the upper fuselage.

The leading gear flaps about the lower fuselage member with the shock absorbing member attached to the forward wing support. The shock absorber consists of a cushion-

iron spring and hydraulic mechanism similar to that used on the larger Stinson model. It is quite simple in operation, having no metering pins or valves. There is an oil filled cylinder with a piston operating inside. The piston has several small holes and ring grooves in it, so that when the lead is applied, the oil is forced through these holes, absorbing the springs. There are two of them, of the bell type. They provide for a travel of 5 1/2 in. so that there is a total vertical travel of the shock absorber of 7 1/2 in., two inches through oil alone and 5 1/2 in. through oil and spring. The oil behind the piston struts is released. The two flaps which are set into the fuselage structure equipment includes 20 in. by 4 in. wheels, though for smaller fields 20 in. by 5 in. tires can be provided. The wheels are fitted with internal expanding brakes actuated by bell cranks and cables and operated from the rubber pedals. The axle tube of the leading gear is attached to the diagonal struts by a steel sleeve welded to the axle and welded to the diagonal strut, thereby eliminating walking on the axle itself.

Wheel Mounted Behind Tail Fins

The tail wheel, which consists of a small disc wheel with a 12 in. by 3 in. pneumatic tire, is mounted on a shock absorber in much the same manner as the front wheels. The hydraulic leg is of similar design to the front wheel shock absorber tube. The tail wheel is connected to the rubber by a diagonal brace which is directly accessible without disturbing the pilot in the usual side seating bumps. The wheel is mounted behind the tail post. It moves almost vertically and has no tendency to reverse itself. This is said to prevent shimmying which is present in some designs. The axle wheel enables the plane to be maneuvered on the ground in high side work without the use of a ground crewman. It is also very saving on the airport layout. As the wheel is mounted behind the tail post the rubber is cut out to provide for the action of the wheel.

The tail surfaces and ailerons are constructed of welded steel tubing. The spar members of the tail surfaces are 1



The Stinson "Junior" owned by the Union Trust Co., Detroit, Mich., on display at the Detroit Show.



Front view of the Stinson "Junior" powered with a Warner "Search" engine.

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The Detroit Air-Olympics

Plans Nearing Completion for the Holding of the National Air Tour, Gordon Bennett Race and National Model Contest

By JOHN T. NEVILL

WITH A "pathfinder" plane now making a circuit of 5,000 mi. per day to make this year the fourth annual National Air Tour, Earl W. Rietz, assistant manager of the tour, and the Aero-Model Society Committee of the Detroit Board of Commerce, sponsors of the event, are busy preparing for the classic.

The tour this year will become a major feature of the Detroit Air-Olympics, a name recently chosen for the air national and international carnival events to be held in Detroit on June 30. The other two events are the Gordon Bennett International Balloon Race, and the Boys' National Model Airplane Contest. The touring planes, taking off in the morning, will open the Air-Olympics and the balloon race will close the program, with the model plane contest, and demonstration of gliders sandwiched in between the two. All of the events are to take place at Ford Airport.

The pathfinder plane, carrying Ray Cooper, tour manager, Ray Collins, official traveling referee, and three Detroit newspapermen, took off from Ford Airport Monday, May 11. Cooper planned to be back in Detroit a week or five before the beginning of field tests preparatory to the tour.

The Proposed Route

On the preliminary flight over the proposed route, the tour manager expects to visit the following cities: Indianapolis, St. Louis, Springfield (Mo.), Wichita, Tulsa, Oklahoma City, Fort Worth, Waco, San Antonio, Henderson (Tex.), El Paso, Tames, Phoenix, San Diego, Los Angeles, Fresno, San Francisco, Oakland, Modesto, Portland, Reno (Neb.), Spokane, Missoula (Mont.), Great Falls, Glasgow (Mont.), Great Falls (Idaho), Fargo, St. Paul, Wausau (Wis.), Milwaukee, Chicago, and Seattle. The cities will be visited in the order named and all of them, at the present writing, are expected to be included in the tour tour.

Past indications are that the tour this year will be larger in every way than any of the previous ones. It will carry the complete plan in approximately 24 cities in 31 states, and will require at least 25 days to complete. Two day stops are planned for several of the stopovers.

Rules and entry blanks for the tour have been mailed to prospective competitors. Manager Cooper said, and the first entries are expected to be in this week.

States Indians have already been entered in the Bennett race, which event will be the second most important of the international classic was established. Four nations, France, Belgium, Germany, and the United States, have entered three men each, with England, Switzerland, Argentina, and Denmark have notified the National Aero-Model Association that they will enter one balloon each. This year marks the first time Argentina and Denmark have competed in the Bennett race.

The Detroit Balloon Club, in whose membership five men are holders of 22 of the 44 world balloon records, recently held a dinner-cum-dance at the Detroit Athletic Club and raised funds for their balloon to be entered from Detroit in the



The start of the 1931 Gordon Bennett International Balloon Race. The "Duo", the balloon entry can be seen just below the ground.

National Balloon Race—elimination contest for the international race—scheduled to be staged in Pittsburgh May 18. The winner of the national race will represent the United States in the Bennett race, one month later. Photo came were Edward J. Hill of the Aero-Model Development Committee of last year's Bennett race, William Maylor, chief engineer of the Stinson Aircraft Corp., Dr. George M. Lohr, Detroit physician and nationally known balloonist, and George H. Brown. Hill will pilot a bag entered by the People's Outfitter Co., owner of the winning bag last year, and will have Arthur O. Schuman, his companion as the prize winning flight, as aide. Henson, pilot of a balloon sponsored by the Detroit Chamber of Commerce, will carry Ernest Johnson, as aide. Aides for the remaining two Detroit entries, both of which are owned by the Detroit Balloon Club and will be sponsored by the Detroit News, have not been named.

Judge Harry S. Hallert of Wayne county probate court, president of the Detroit Balloon Club, and ardent balloonist enthusiast has been appointed by the Aircraft Events Committee of the Board of Commerce as chairman of the balloon

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Reducing the Cost of Fire Insurance

By WILLIS PARKER

RATES CHARGED for various types of insurance sought by the airplane manufacturers are so high that any little saving effected is noticeable in the overhead expense of the industry in paying, but growing. Many of the small firms are at expanding their facilities, remodeling or rebuilding their factories, and they should, at those times, consider their insurance needs.

By making a few changes in their present factory building and incorporating many improvements in the new and structures being added, the Swiflow Airplane Co. of Wichita, Kan., has been able to reduce its fire insurance rates from \$100 a hundred to \$150—quite a saving in the cost of a job, few years or more. And, to better describe the company's new plant, we may well picture it from the fire insurance angle.

At the outset the new buildings are of brick, steel, concrete construction. The first building was rock, but, in an effort to add more structure architectural features, a first addition was made of brick and concrete. There had to be a way to reduce the fire hazard and also the insurance rate. For example the ceilings of the present office and warehouse are covered with wall board and other such materials to make the rough beams and boards in the construction. The walls were shrouded, also, and there was sufficient of such material to classify the building as "hazard" instead of "good build". By removing sufficient shoring and ceiling so that not more than 5 per cent. of the total building was added, a reduction of ten cents on each hundred dollars insurance was obtained.

Rate Reduced 40 Cents a Hundred

The handling of "dope" and wood working in the same building with the necessary or go working together a hazard that the insurance companies consider worth punishing. Eliminating all woodwork, both hard and power, from this building reduced the rate 40 cents a hundred. The discontinuance of all painting and the storage of dope in the building saved another \$1.50 to be taken off the rate.

Elimination of the storage of gasoline for any purpose whatever brought a cut of 30 cents. The discontinuance of the use of gas turbines and using only gas-turbine, suitable or external gas portable devices would give a 12 cent reduction, a re-wiring of the building for electricity would reduce the rate 15 cents.

If the company will not permit the storage of extra acetylene or oxygen tanks other than those actually necessary to wiring the process in the building a reduction of 15 cents will be granted.

A reduction of 15 cents also is given for replacing of any gas or break type electric motor with "straight induction" or totally enclosed types.

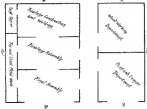
So all in all, many more, metal boilers for workmen's city clothing gives the firm the advantage of 5 cents a hundred reduction in the rate. Another five cent reduction comes from

the installation of standard metal waste cans with self-closing latched lids in all locations where oily rags are used or accumulated.

Concrete floors instead of wood reduce the rate, so whenever possible wood floors in the old factory will be replaced with concrete and if it is possible to make 30 per cent of the floor area concrete it will reduce the rate 10 per cent. A one cent reduction will be granted when the placing of above pipes



Layout of the factory units of the Swiflow Airplane Co., Wichita, Kan.



through combustible roofs and the openings thus commenced are permanently closed, a discontinuance.

The recommendations of the fire insurance agents will be carried out in remodeling of the old factory building, says Victor Rose, general manager of the company, as well the recommendations made with the erection of the new buildings or units to the factory.

The wood working and dope departments are being housed in a building separated from the old factory building by a 20 ft. driveway. Had a fire wall will separate the two departments and extend 15 ft. above the roof. This will make it necessary to make the material outside of the building when shifting it from one department to another, but with wide doors, a narrow runway, which can be covered with a light canopy which is not attached with the buildings and thus prevent the material and the workmen in bed weather, it will

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Rules of the 1928 National Air Tour

THE EDSEL B. FORD Reliability Trophy, donated by Edsel B. Ford, shall be competed for annually by commercial airplane manufacturers. Entrants must show to the satisfaction of the Contest Committee of the N.A.A. that they are bona-fide manufacturers.

(2) The contest shall be in the nature of a symmetrical tour from city to city and shall be conducted after a course and under such rules as may be prescribed from time to time by the Contest Committee of the National Aeronautic Association.

(3) The Edsel B. Ford Trophy shall be awarded each year, to the entrant of the winning airplane, subject to the rules and conditions hereafter set out, who shall be entitled to possession of the Trophy until one month prior to the next succeeding contest at which time he shall deliver the Trophy to the Contest Committee of the National Aeronautic Association. A suitable bond for the proper care and return of the Trophy shall be required by the National Aeronautic Association from each person or organization into whose possession the Trophy may be delivered.

(4) The Edsel B. Ford Reliability Trophy shall become the permanent property of any manufacturer who shall win it on three consecutive occasions, and shall not, without his approval be again contested for.

(5) In order to stimulate entries among civilian contestants and aviation pilots, both in the United States and abroad, government-owned planes will not compete in this Tour.

(6) For the purpose of this Contest, at least four entrants shall be represented with a minimum number of entrants not less than ten. In the event that total entry list does not aggregate ten contestants representing at least four entrants no contest shall be held.

Rules and Conditions for 1928

(7) Each airplane entered in the National Air Tour shall be capable of a speed in excess of eighty (80) miles per hour with a full combat load, and shall be required to demonstrate this speed before being allowed to compete in the Tour.

(8) No airplane shall be eligible to compete in the National Air Tour of 1928 unless an Approved Type Certificate has been issued by the Department of Commerce of the United States for that particular type of airplane.

(9) An exception to this may be granted to entrants who have not yet received Approved Type Certificates from such Department, provided they have filed application for such certificates with the Department together with the supporting data required by the Department, as far before June 15, 1928, thus enabling the Department to determine by a suitable check thereof, together with an inspection and flight test of the airplane received, whether it is in order to issue an experimental license for the purpose of the Tour. The exceptions of such entries shall be optional with the Tour management.

Content Load.

The content load is the load which the entrant desires to

put in the plane providing the following load conditions are met to the satisfaction of the Contest Committee:

(a)—Where no circumstances shall the content load exceed the maximum useful load permitted by the Department of Commerce under the approved type certificates issued for the plane.

(b)—This content load may consist of passengers, fuel and oil and useful ballast, or both, and must include the weight of the pilot and the weight of gasoline with the tanks full. Tires, spare parts, cockpit covers may be included in the content load when properly sealed and marked.

(c)—For the purpose of the check and suitable maximum speed tests, the content load will consist of passengers



Edsel B. Ford (right) and party standing in front of the Winner-Detroit which he flew to first place in the 1927 Tour.

and ballast, or ballast in addition to full tanks of gasoline and in sufficient quantity to bring the content load of the plane up to maximum useful load permitted by the Department of Commerce for such step. After the check, action, and maximum speed tests have been completed, the entrant may reduce his useful load to 75 per cent. of the maximum permitted by the Department of Commerce and still be given credit for 100 per cent. in the formula. Reasoning that justice is a part of the useful load specified under the regulations of the Department of Commerce and that the weight of pressure varies with the distance traveled, each plane will be required to leave each control with full gasoline tanks. The same will apply to oil tanks except that a margin of 25 per cent. of the capacity of oil tanks will be allowed for consumption. Upon arrival at each control, each entrant will be required to account for a full 75 per cent. of the maximum useful load permitted by the Department of Commerce on

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N.A.C.A. Progress

Industry Delegates Witness N.A.C.A. Equipment Demonstrations at Third Annual Conference Held at Langley Field

COOPERATION WITH the aircraft industry and the expression of a desire to help commercial manufacturers to solve their problems was the subject of many of the Third Annual Conference of the National Advisory Committee for Aeronautics which was held at Langley Field, Va. on May 15. About 95 members of the industry attended, most of them being the heads of aircraft companies or chief engineers. That is a greater number than has attended any previous conference, and though there was a preponderance of military manufacturers, many of the more progressive commercial manufacturers were representatives.

As it will be remembered, the National Advisory Committee for Aeronautics was started in 1915 in order that the United States might have an independent body to which the government, aviation, and especially the Army and Navy, might go for technical information about modern aeronautics. Information was to be published and fundamental research of aeronautics problems was to be undertaken. The organization proved of such value that the appropriations were gradually increased, it now has, including the Washington staff, a total personnel of about 250, and received an appropriation of \$400,000 last year.

At first, work was done mostly for the Army and Navy, but as facilities have increased, the aircraft industry has become more and more interested in what is being done. Information on the data gathered and on the actual experimental work at Langley Field is sent out in a series of technical bulletins which are of great value to the engineers of the country. The conference of the industry and the N.A.C.A. have brought about closer collaboration of the technical work being done and the practical needs of the industry and of the military. With the growth of the need for better expression of commercial aircraft, it is hoped that the design of civilian planes will formulate their needs and problems, so that they may more fully profit by the scientific facilities available.

The present experimental plant at Langley Field is probably the complete as any in the world. To the outside, the

most striking piece of apparatus in the wind tunnel which blew a jet of air 20 ft. across a maximum velocity of 120 m.p.h. The eight bladed propeller which operates this air is driven by two 1,800 h.p. Diesel engines. The size and speed of the tunnel enable tests to be made on full sized airplanes with their engine running. In an ordinary wind tunnel it is not possible to measure the effect of the differences of the



Mouth of the 20 ft. N.A.C.A. wind tunnel showing method of measuring actual flight conditions.

glass's propeller and its reaction on the fuselage and wings. This is an important factor in design, but at present little is known about it. A Whittaker engine valve arrangement is now being tested in the tunnel. A variety of different settings are being tried and electrical thermometers are placed on various parts of the engine and cylinders. It is hoped that interesting results will be obtained, not only as to the best form of cooling from an aerodynamic standpoint, but

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Group picture of delegates who attended the Third Annual Conference of the National Advisory Committee for Aeronautics.

Fuselage Analysis

Stress Analysis of Commercial Aircraft, Chapter Number Twelve

By PROFESSOR ALEXANDER KLEMIN

David Hughes, Editor of AVIATION

And GEORGE F. TITERTON

Chief of the Bureau of Aeronautics, Navy Department

IT IS the common practice to solve the fuselage truss graphically. This method is probably less lengthy than the analytical method and has the added advantage of providing a check on the work and of readily perceiving the magnitude of the stresses. Its disadvantages are that some time must be taken to set up the truss accurately, the solution and especially accurate drawings of stress and angles are required if the diagram is to close. Before starting to analyze we must be quite sure that the forces acting on the truss are in equilibrium. If the forces are not in perfect equilibrium the diagram will not close and the analysis must be repeated.

As explained in the last part of Chapter I, in doing graphical work it is advisable to construct the force polygon first—that is, a polygon of all the external forces acting on the truss. If this diagram closes we are sure that our X and Y force balances and the moment equilibrium must be checked independently. It should also be borne in mind that if we take the members in reverse clockwise about any point, then stresses which we draw on our polygon toward the joint are in compression, and those looking away from the joint are in tension. A letter should be put on each side of every member and force of the truss and the letter is the polygon named as they are drawn.

The three polygons must of course be drawn to a common scale. For a 2,000 lb. plane a scale of about 300 lb. to the inch will give a fairly large size polygon. The larger the polygon the greater is the accuracy but large polygons are somewhat awkward and errors other than measurement creep in. The polygons to be published as this series have been solved with scales from 250 to 500 lb. per in. but in publishing the printer will reduce the size to fit the page and so the scale will no longer hold true. To determine the scale of the printed polygons measure one line of known magnitude and divide this magnitude by the measurement in inches.

In the first three conditions that will be dealt with, namely, maximum stabilizer and elevator load, maximum forward and rear load, and wing over, it is not necessary to analyze the entire truss of the fuselage. These three conditions are fairly simple and enable one to become familiar with the methods of graphical analysis.

Maximum Stabilizer and Elevator Load

In this condition a required load per square foot must be applied at the tail post and the rear truss of the fuselage analyzed as far forward as the rear of the cabin. In the previous Chapter we found that 30 lb. per sq. ft. in the load required on one horizontal sq. ft. of tail surface. In Fig. 30 of Chapter 8 the area of the stabilizer and elevator is given as 34 sq. ft. The load on the tail post is then $30 \times 34 = 1020$ lb. Half this load has been applied at the tail post and a graphical solution made in Fig. 72. Only half the load was used because only one side of the fuselage is being analyzed.

The upward reaction of the fuselage is being analyzed. The required reaction of the fuselage is being analyzed. The required reaction of the fuselage is being analyzed.

Assuming that a represents all the stress in one side of the internal force of 1020 lb. and b represents all the stress on the other side. This force is then represented by $a-b$ which is a descriptive in the polygons. It is to be noted that the tail post and the rear truss of the lower longons both have zero load. This is due to the fact that the truss is complete without these two members so far as this condition is concerned. Let us follow through the first panel point where to see if we understand the method. We have our external force $a-b$ acting vertically downward with a force of 1020 lb. A line $a-b$ is drawn below the polygon to a definite scale and parallel to the line $a-b$ acting on the truss. Proceeding clockwise about the panel point we come to member $b-a$ (tail post) which we just said was zero and therefore has a zero value. A line is drawn through point a of the poly-



gon parallel to $a-b$ in the truss and perpendicular to length. The tail and last member of this panel point is $b-a$. Through the point a in the polygon a line is drawn parallel to $a-b$ until it intersects the line $a-b$ which was drawn last. The point of intersection is the point a . Now by measuring $a-b$ and multiplying by the scale to which $a-b$ was applied, we can obtain the stress in this member.

The force $a-b$ is acting downward and in one polygon it is so labeled that when we read $a-b$ we go from top to bottom—that is, in the direction in which the force is acting. If we now read $a-b$ on the polygon we go from lower left to upper right. This is acting right towards the panel point we just analyzed. The force $a-b$ then is pushing toward the panel point and the force is in compression. A component force of 1760 lb. was obtained by measurement and the b label in the truss and indicated by -1760 . Minor stresses in compression and also reactions tension. Reading the force on the polygon the stress from right to left is away from the panel point. It is therefore in tension. The magnitude of

member $a-b$ is indicated by -1430 on the truss. The rest of the truss is analyzed in the same manner until the loads become very small and are no longer evident.

Maximum Forward and Rudder Load

In this condition it is assumed that the vertical tail surfaces are a half per square foot of three-quarters the magnitude of the horizontal tail surfaces. The horizontal tail is loading at 30 lb. per sq. ft. and so the load on the vertical tail is $30 \times 34 = 1020$ lb. per sq. ft. The area of the vertical tail is given as 34 sq. ft. in Fig. 30 of Chapter 8. The total load is then $1020 \times 34 = 34680$ lb. This load is acting at the rear of the fuselage and the vertical tail surfaces. This point is usually assumed as the geometrical center of the vertical tail surfaces. A point is selected that has as much area above it as below, and a point is drawn as in Fig. 73. Unless the rudder is a fairly conventional one, the force and its distribution of area is required and the point placed somewhere along the vertical tail of the proper height to give equal areas above and below. In our case this point is about three inches above the upper longon.

The force of 34680 lb. acting three inches above the upper longon puts the fuselage structure in tension. It is rather difficult to analyze far forward and so the loads are made a component in tension. In the lower right hand corner of Fig. 73 is a little diagram showing the tail or rudder part of the plane. The force of 34680 lb. is acting three inches above the upper longon which is two is 18 in above the lower longon. If we take moments about the lower longon we will obtain the force necessary at the upper longon to maintain equilibrium. Thus $34680 \times 18 = 30/18 = 34680$ lb. acting at the upper longon. For equilibrium assume a force of 4975 lb. must be applied at the lower longon. If the upper truss is now analyzed for a load of 1020 lb. acting at the rear post of the longon, stresses will be induced in the members which will be perfectly accurate. Similar members in the lower truss will be assumed to have the same stresses as the members of the upper truss.

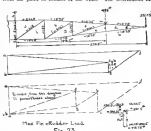
In Fig. 73 the upper truss of the fuselage has been analyzed as far forward as the rear post of the wing. The full-line polygon is drawn in this condition. In the lower right hand corner of member $a-b$, $d-e$, $a-f$, $g-h$, $i-j$ will carry any load. This is due to the fact that the two longons with member $a-b$ form a triangle which is a rigid structure in itself and does not require any internal bracing. Now however if the internal bracing was omitted the length of the longons would be so long they very heavy when would be required to take the loads. To break up the column length of the longons and to make the rear portion of the fuselage more rigid other members are inserted. In the full-line polygon these members are all represented by a point which indicates they carry no load. When the longons change in size the members $b-c$, $d-e$, $f-g$, $h-i$ will be stressed as shown in the full-line polygon by $b-c$, $d-e$, $f-g$, $h-i$.

To stress some idea of the loads in the lower members which show it to be in the full-line polygon the following method has been employed. The line of the rear post is taken as a reference point. It is desired to place one panel point in the rear of the tail and will have the same effect as the reference point as a load of 34125 lb. at the tail. Taking moments about the reference point this load must be $34125 \times 20/20 = 34125 \times 10/20 = 1706$ lb. This load is applied at the rear panel point and indicated by a dashed arrow in Fig. 73. The load is shown in the lower polygon as indicated by the dashed polygon. Loads are obtained on $i-h$ and $h-g$. These members probably do not truly carry these loads but they are at least an indication upon which the design of the members can be based. These members are very small and $h-g$ is usually possible to use as the minimum tail and page of taking that adding consideration

will allow. The Commerce Department specifies minimum area of taking that may be used. These will be given later.

Wing Over Condition

In this condition the plane is assumed to be rolling as shown in Fig. 74. The weight of the plane times the three point landing load factor is acting vertically downward through the center of gravity. In our plane this load is $2000 \times 3.0 = 6000$ lb. This is only half the total load of the plane but our analysis is being made for only one side of the fuselage and this will give the true loads on the members. The line of action of this load acts the ground 30 in. from the point of contact of the engine banner and 30 in. from the point of contact of the wheel. The distribution of



the load between these two points is then: $7000 \times 20/50 = 2800$ lb. at the wheel; $7000 \times 30/50 = 4200$ lb. at the engine mount.

The chief purpose of the analysis of this condition is to make the forward part of the fuselage strong enough to withstand the wing over loads. It is desired that if anything is to fail in the landing gear on this, the plane will assume a horizontal attitude. With this in view the landing gear is not analyzed for this condition but only the forward part of the fuselage. The load of 3000 lb. acting at the wheel is then carried through the landing gear but left at its wheel. The load of 4200 lb. is carried through the front portion of the fuselage structure.

The two upper members of the engine mount do not take any load as the structure is complete for this condition without these members. The loads in the other members of the fuselage and engine mount are rather severe and it is noted that the engine mount is designed by these loads.

Reference Diagram

At this stage of the analysis it is essential that we know the exact distribution of air weight. With this in view we have constructed a reference diagram in Fig. 25. The plane is placed exactly as called for in the balance table which was published in Chapter 9 in analyzing the landing gear. The horizontal reference line, the front line of the rear propeller legs is located on the diagram; the vertical reference line, the thrust line is also drawn on the diagram. All Q forces are given with reference to these datum lines.

In dealing upon the order of gravity position of any part

Last Minute Briefs

Alexander "Tony" Thompson has incorporated the Thompson Flying Service at Bell Lake City, Utah.

Jim Burke has taken the agency for Travel Air planes in half a dozen Kansas counties. Burke has a hangar completed at Newton, Kan., his headquarters.

Bert Wright, engine mechanic, has taken the agency for Swallow planes for 11 counties in Central Kansas. He now has a hangar under construction at Newton.

The Co-Operative Flying Club has been formed in the western part of Chicago. Two Boeing Air Transport Co. pilots have been secured to give instruction to members in the club's newly-purchased Travel Air plane.

The factors of the American Eagle Corp. at Kansas City, Mo., is being enlarged with a 30 by 100 ft. building to be used as a wing department. The company's plane production is being increased from eight to 12 planes a week.

Forbes Barker, commercial agent with the Government's Federal Bureau of Surveys and Domestic Commerce, recently was transferred to Washington to become chief of the aircraft section. Transportation Division, Department of Commerce. Barker served as pilot with the British forces during the War.

Dr. Hugo Junkers has returned to New York after having made a tour which included Philadelphia, Washington, and several Midwestern cities. The Junkers Corporation head plans to stay in this country several more weeks.

Leo Haslam Harris, patent expert engineer who has specialized in this class of work since 1910, is now located at Brightwaters, L. I. He is prepared to handle all phases of airport engineering, he states.

The L. C. Armstrong Co., custom body builders, plan to start production on airplane fuselages, wings, and other within a short time. The company, which is located at 1635 El Centro Ave., Hollywood, Calif., will be in a position to manufacture these products for Eastern manufacturers.

The Diesel Supercharger Aeronautical Laboratory of the Massachusetts Institute of Technology, Boston, will be formally dedicated on Monday afternoon, June 4.

Edward F. Bellini, president of the Wayne Oil Co. of Detroit, owner and pilot with William B. Brock of the Fields of Detroit, has entered a Bellanca monoplane in the first official entry in the 1932 National Air Show, which begins from Ford Airport June 30. William Brock will pilot the Bellanca.

Ray Fieberly (newspaper) transport monoplane leaving Kay Cooper and three passengers now has left Detroit on a 6:00 a. m. paddling trip to map the route of the 1932 National Air Tour.

The Model 447 C Fairchild Commion has successfully passed the U. S. Army Air Corps 54 ft. test. Passage of this test is a requisite for Air Corps recognition as a standard production engine.

The Boushon Aircraft Corp. of Hillsboro, R. I., has completed its first biplane, the "Kitty Hawk", powered with a seven cylinder Ryan Remont engine. A test of this big bird will shortly be made in the plane.

Tests and calibration of a new ground speed indicator has been begun by Dr. Adam Hoffman, dean of the School of Engineering Division at the University of Dayton, O. W. C. Rogers, formerly in the Air Service, directed the device.

Capt. George White in conducting his experimental flight in his aircraft at St. Augustine, Fla., despite a head fall from an altitude of five feet which damaged the right wing of his biplane craft.

A. V. Thaddeus, vice president of the Thomas Metal Works Corp. of San Francisco, announces that the company is working on a new four phase, full cantilever wing, all metal monoplane. It is expected that two of the planes will be ready by July 1.

W. B. Allen has become field manager and F. J. Ogan, assistant of the Steeple Airline, Inc., Eagle Rock Airplane a Western Kansas and Eastern Nebraska.

Levi Clifton C. Haldeman has been appointed flight manager and general director of the flying school of the Robinson Flying Service, Inc., at Lambert-St. Louis Field, St. Louis, a graduate of the Army Flying School, is a transport pilot.

Levi John Jolinger has been appointed an instructor of the Robinson Flying Service, Lambert-St. Louis Field, also an Army Flying School graduate.

General Noble has made a flight to the North Pole from land in his dirigible Italia. The Italian dirigible, while it had its trip because of fog, returned to his headquarters where he is now preparing for a flight to the Pole itself.

G. M. Bellanca is to build a monoplane powered with a Wasp engine placed in tandem, it is reported, for "Staro" Stoltz, former test pilot and Harmon Trophy winner. That it is successful, will attempt a trans-Pacific flight.

Tankers of air mail from the Newark, N. J. field, to the New York Post Office will take 35 min. as compared with the two hour land from Hedges Field, at New Brunswick, which is now used. The Newark field will be operated beginning August 1.

A new aerial camera has been developed by the Fairchild Aerial Camera Corp. of New York in collaboration with Army experimental section at Wright Field, Dayton, O. The camera will be described in an early issue of AVIATION.

The Aero Coupe, a convertible cabin biplane, recently produced by the Aero Craft Manufacturing Co. of Detroit has been flight tested. A complete technical description of the plane will appear in an early issue.

Donaldson Givens and Joseph Lebrun, around the world fliers, have begun a tour of Europe. Following their return to Paris, they will prepare for a flight to New York, it is said.

The Works Air Craft Corp. of Milwaukee, Wis., has taken the agency for the Monocoupe plane. The Monocoupe is made by the Moss-Albion, Inc., of Moline, Ill.

G. M. Bellanca Completes Large Hornet

Powered Sesquiplane for Cesare Sabelbi
GISEPPE M. BELLANCA has designed and completed the hornet plane, a sesquiplane powered with a single 300 hp Pratt & Whitney engine. The Model K Bellanca has been completed for Cesare Sabelbi, an Italian World War ace, who is completing a flight to Rome and return accompanied by Roger C. Williams, pilot, and Patsy Benda, navigator and radio operator.

The plane has many original features including a retractable landing gear, an auxiliary wing or lift strut of nearly six



The Bellanca Model K, long distance type powered with a Pratt & Whitney Hornet engine. Photo was taken during preliminary flights at Hedges Field, New Brunswick, N. J.

feet ahead of the root and two foot chord at a point slightly below the upper wing and also a portion of the wing structure of metal covered with sheet duralumin making this part of the wing a girder beam. The overall span of the model is 50 ft., it weighs 4,000 lb. empty with tanks, instruments, machine, etc. The gasoline capacity is 210 gal. which should give a range of approximately 5,700 mi. The plane was recently tested at Hedges Field, New Brunswick, N. J., by Roger C. Smith, test pilot for the Bellanca Aircraft Corp. Word from Hedges, N.J., states that the project has been reported there by a group of Italian-Americans. The plane is said to have cost \$15,000, and it is stated that the Bellanca Association of Commerce has been asked to raise an additional \$15,000 to cover the other expenses of the flight. The Motor and the association of commerce are very much interested in a flight of the kind and a committee has been organized to investigate the proposition. The committee is composed of Horrie D. Bell, J. W. Tottle, Robert Munnison, Jack Boushous, and David Louis Hopkins. All are prominent Bellancians.

A complete description of the Model K Bellanca will appear in an early issue of AVIATION.

Keystone Company is Awarded Government Contract for 35 "Panther" Bombing Planes

A CONTRACT for 35 Keystone "Panther" bombing planes costing approximately \$1,200,000 as full has been awarded by the U. S. Government to the Keystone Aircraft Corp. of Erie, Pa., as a result of an announcement by Edgar M. G. president. This is reported to be one of the largest contracts awarded an aeronautical firm for the manufacture of bombers planes. The Panther was one of the 16 types designed according to Air Corps specifications and selected by manufacturers to the Government for evaluation tests at Wright Field, Dayton, O.

Apparently half of the machine required for these planes will be 325 hp. Wright "Cyclone". The remainder of the engine will be "Hercules" furnished by the Pratt & Whitney Corp. of Hartford, Conn. An interesting comparison of these two engine types of air cooled engines in actual service will

be afforded. The Wright Cyclone recently reached the P.N. 12 Navy airplane in which it was world's airplane endurance record of over 36 hr. at Philadelphia.

The Keystone Aircraft Corp. has contracted to deliver the first Panther by next October, with the full order to be completed and delivered by July 1933.

Price of Looming Cabin Amphibian Plane Is Increased from \$24,700 to \$26,700

AN INCREASE in price of the Looming Cabin Amphibian plane has been announced by the Looming Aeronautical Engineering Corp. of New York City. The plane will formerly sell for \$25,700. The former price was \$24,700.

The Looming Cabin Amphibian is the first plane built by the company for commercial use and is a direct development from the Looming O-8, the standard service type now in production for the U. S. Navy. The plane is powered with the 125 hp. Wasp engine.

A number of Looming Cabin Amphibians are now under construction, according to Berkeley Haven, sales manager of the company. The Canadian Trans-Continental Airways Co., Haven asserts, has purchased one of these planes for mail and passenger service to the maritime provinces of the Dominion.

REVIEWS

The first Department aviation book that will be New York City is a copy of the book entitled "The Aeronautical Engineer's Handbook" published by the National Aeronautics Association, Washington, D. C.

A B C of Flight by W. Laurence LePage, published by John Wiley and Sons, Inc., New York City, 141 pages, price \$1.00.

This book, with a foreword by the Assistant Secretary of the Navy for Aeronautics, Edward E. Street, presents in a clear, simple language the fundamental principles of the science and art of flight. It assumes no previous knowledge of the subject or of any knowledge of mathematics or physics beyond that of every-day observation. Illustrated with a number of photographs and sketches, A B C of Flight should prove an excellent supplement to a ground school course or the new textbook known as the fundamentals of flight. Both the airplane and airplane engine are described and, in addition, there is a chapter devoted to flying instruction. It makes no pretense at teaching the reader to fly, but is intended as to be of material effort in one taking actual flying instruction.

Aircraft Engine Instruction edited by A. L. Byke, published by The Goodheart-Willcox Co., Chicago, Ill., 322 pages, price \$3.00.

In the Aircraft Engine Instruction, Byke has compiled a wealth of information on the construction, maintenance and operation of aircraft engines and accessories. This handbook gives the data on the best known aircraft engines, 60 in all in the list of modern aircraft engines in a most thorough manner. It is well illustrated with photographs of the various engines bringing out structural details and methods of adjustment and assembly. In addition, there are a number of line sketches, many of which are in color. Each of the engines is described in detail, while a complete chapter on the design is devoted to various engines which are now in use. Approximately one-third of the book is taken up with accessories, such as carburetors, exhausters, starters, instruments, controls, etc.

Many Civic Leaders in the San Francisco

Area Boost Citizens' Aviation Association

TWENTY-TWO CIVIC leaders and East Bay aviation enthusiasts have endorsed and accepted membership in the Citizens' Aviation Association, San Francisco, according to Frank W. Epperson of Oakland, president. These include: Guy Tamm, E. B. Field, Max Horvitz, May Livingston Irving, William B. Parker, Emil Frutkin, E. B. Austin, F. M. Buhler, W. E. La Follis, Capt. George D. Evans, C. L. Gilbert, Joseph D. Gardner, Henry W. Young, Capt. W. H. Voornoy, H. B. Aschbach, Herman C. Ferguson, Philip M. Fisher, Capt. D. E. Doty, J. F. Long, A. E. Wilde, John L. Bursey, J. B. John, L. E. Hunsman, H. E. Lebold, W. G. Bullock, H. H. Schuler, Merle J. Conn, C. A. Aikin, A. W. Neeking, W. E. Feltner, and G. E. Lashburn. The prime object of the organization is to promote public interest in aviation, and to encourage private enterprise in that line.

David Bell Company Issues a New Catalog

Of Its Automatic Screw Machine Products

DAVID BELL Co., Inc., of Buffalo, N. Y., recently issued a new catalog of its automatic screw machine products. Part of this catalog is devoted to the aircraft products manufactured by the company, which include propeller hub pulleys, valve pulley parts and assembly, exhaust bracket and valve drive pins, strut assembly parts, and special bolts of all descriptions.

Bell products are made from cold-chamber or hot rolled special screw stock, from alloy or special steels, or from other similar materials. They can be heat treated, heat-proofed, or plated.

New Tri-State Airlines, Inc., at Sioux

City is Ryan and American Eagle Dealer

TRI-STATE AIRLINES, Inc., has been organized to operate at Sioux City, Ia., at the Eisenhower Airport. One hangar is already built at the field and another will shortly be constructed.

The company, which is American Eagle and Ryan plane distributor, will use the former type planes for instruction in the flying school scheduled to open June 1 and a Ryan plane for general transportation and taxi work. Boyd Miller, president of the company, and James Barwick, vice president and chief pilot, are going to San Diego, Calif., to bring back the Ryan plane ordered. Frank Mahall is business manager of the company, while Dale Jones is secretary and treasurer.

Fuselage Analysis

Continued from page 1327

builder does a little judgment in all that is needed. For irregularly shaped tanks and such things the center is taken, for pilot and passengers somewhere about the stomach is a good location, for wings 48 per cent of the chord back of the leading edge, propellers have their center of gravity right on the thrust line and about in the center of the hub. Engine manufacturers usually specify the location on their blueprints of the engine; the fuselage location depends upon the shape and type but as a general rule it is well to remember that the fuselage structure is such bearing up forward where most of the weight acts. When there is some doubt as to the exact location of the center of gravity of an item and one guess is as good as another, select the most convenient location as it will simplify later work. For instance, in locating the center of gravity of the left group plane it at the tail-

post of no better location for it is known. This will simplify the work in making up the panel load table which is the next thing to be done.

Distribution of Panel Loads

In the analysis of the four remaining major members, namely, high and low incidence, level landing and three-point landing, it is necessary that all the weights of the item be applied at the panel points. These panel points are designated by astral numbers in Fig. 75. The weights are ap-



Hooping Over

Fig. 74.

plied among the panel points which surround the particular weight. For example, referring to Fig. 70, the passenger weight of 200 lb. lies between panel points 5, 6, and 7; it must therefore be distributed among these points.

In splitting up weights between panel points it is essential that the moments of all loads at the panel points about a reference line be the same as the original moments of the weights themselves. If this is not the case that means a center of gravity of the plane has moved and is the analysis of the four remaining conditions the polygons will not close and we will have no check on our work. It therefore pays to take great pains in determining our panel loads.

A table of panel loads has been made up for our plane in accordance with the balance diagram of Fig. 75. The list



Balance Diagram

Fig. 75

step is making up such a table is to list all the panel points and give the horizontal and vertical axes of each point from line to the datum line. These axes are to be measured very carefully off a line diagram of the fuselage drawn in scale. The next step is to list the balance table as it was given in Chapter 16. This has been done in the first six columns of our panel load table. We are now ready to analyze the panel loads.

The propeller and the engine are not forward of any of our panel points and consequently can not be distributed between them. These weights may be placed at panel points 10 and 11, with a horizontal couple at these points sufficient to balance the moment of these weights about their datum

May 27, 1928



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Marysville Michigan

PANEL POINT LOADS									
Panel Point	1	2	3	4	5	6	7	8	9
100 lbs.	100	100	100	100	100	100	100	100	100
200 lbs.	200	200	200	200	200	200	200	200	200
300 lbs.	300	300	300	300	300	300	300	300	300
400 lbs.	400	400	400	400	400	400	400	400	400
500 lbs.	500	500	500	500	500	500	500	500	500
600 lbs.	600	600	600	600	600	600	600	600	600
700 lbs.	700	700	700	700	700	700	700	700	700
800 lbs.	800	800	800	800	800	800	800	800	800
900 lbs.	900	900	900	900	900	900	900	900	900
1000 lbs.	1000	1000	1000	1000	1000	1000	1000	1000	1000
1100 lbs.	1100	1100	1100	1100	1100	1100	1100	1100	1100
1200 lbs.	1200	1200	1200	1200	1200	1200	1200	1200	1200
1300 lbs.	1300	1300	1300	1300	1300	1300	1300	1300	1300
1400 lbs.	1400	1400	1400	1400	1400	1400	1400	1400	1400
1500 lbs.	1500	1500	1500	1500	1500	1500	1500	1500	1500
1600 lbs.	1600	1600	1600	1600	1600	1600	1600	1600	1600
1700 lbs.	1700	1700	1700	1700	1700	1700	1700	1700	1700
1800 lbs.	1800	1800	1800	1800	1800	1800	1800	1800	1800
1900 lbs.	1900	1900	1900	1900	1900	1900	1900	1900	1900
2000 lbs.	2000	2000	2000	2000	2000	2000	2000	2000	2000
2100 lbs.	2100	2100	2100	2100	2100	2100	2100	2100	2100
2200 lbs.	2200	2200	2200	2200	2200	2200	2200	2200	2200
2300 lbs.	2300	2300	2300	2300	2300	2300	2300	2300	2300
2400 lbs.	2400	2400	2400	2400	2400	2400	2400	2400	2400
2500 lbs.	2500	2500	2500	2500	2500	2500	2500	2500	2500
2600 lbs.	2600	2600	2600	2600	2600	2600	2600	2600	2600
2700 lbs.	2700	2700	2700	2700	2700	2700	2700	2700	2700
2800 lbs.	2800	2800	2800	2800	2800	2800	2800	2800	2800
2900 lbs.	2900	2900	2900	2900	2900	2900	2900	2900	2900
3000 lbs.	3000	3000	3000	3000	3000	3000	3000	3000	3000
3100 lbs.	3100	3100	3100	3100	3100	3100	3100	3100	3100
3200 lbs.	3200	3200	3200	3200	3200	3200	3200	3200	3200
3300 lbs.	3300	3300	3300	3300	3300	3300	3300	3300	3300
3400 lbs.	3400	3400	3400	3400	3400	3400	3400	3400	3400
3500 lbs.	3500	3500	3500	3500	3500	3500	3500	3500	3500
3600 lbs.	3600	3600	3600	3600	3600	3600	3600	3600	3600
3700 lbs.	3700	3700	3700	3700	3700	3700	3700	3700	3700
3800 lbs.	3800	3800	3800	3800	3800	3800	3800	3800	3800
3900 lbs.	3900	3900	3900	3900	3900	3900	3900	3900	3900
4000 lbs.	4000	4000	4000	4000	4000	4000	4000	4000	4000
4100 lbs.	4100	4100	4100	4100	4100	4100	4100	4100	4100
4200 lbs.	4200	4200	4200	4200	4200	4200	4200	4200	4200
4300 lbs.	4300	4300	4300	4300	4300	4300	4300	4300	4300
4400 lbs.	4400	4400	4400	4400	4400	4400	4400	4400	4400
4500 lbs.	4500	4500	4500	4500	4500	4500	4500	4500	4500
4600 lbs.	4600	4600	4600	4600	4600	4600	4600	4600	4600
4700 lbs.	4700	4700	4700	4700	4700	4700	4700	4700	4700
4800 lbs.	4800	4800	4800	4800	4800	4800	4800	4800	4800
4900 lbs.	4900	4900	4900	4900	4900	4900	4900	4900	4900
5000 lbs.	5000	5000	5000	5000	5000	5000	5000	5000	5000
5100 lbs.	5100	5100	5100	5100	5100	5100	5100	5100	5100
5200 lbs.	5200	5200	5200	5200	5200	5200	5200	5200	5200
5300 lbs.	5300	5300	5300	5300	5300	5300	5300	5300	5300
5400 lbs.	5400	5400	5400	5400	5400	5400	5400	5400	5400
5500 lbs.	5500	5500	5500	5500	5500	5500	5500	5500	5500
5600 lbs.	5600	5600	5600	5600	5600	5600	5600	5600	5600
5700 lbs.	5700	5700	5700	5700	5700	5700	5700	5700	5700
5800 lbs.	5800	5800	5800	5800	5800	5800	5800	5800	5800
5900 lbs.	5900	5900	5900	5900	5900	5900	5900	5900	5900
6000 lbs.	6000	6000	6000	6000	6000	6000	6000	6000	6000
6100 lbs.	6100	6100	6100	6100	6100	6100	6100	6100	6100
6200 lbs.	6200	6200	6200	6200	6200	6200	6200	6200	6200
6300 lbs.	6300	6300	6300	6300	6300	6300	6300	6300	6300
6400 lbs.	6400	6400	6400	6400	6400	6400	6400	6400	6400
6500 lbs.	6500	6500	6500	6500	6500	6500	6500	6500	6500
6600 lbs.	6600	6600	6600	6600	6600	6600	6600	6600	6600
6700 lbs.	6700	6700	6700	6700	6700	6700	6700	6700	6700
6800 lbs.	6800	6800	6800	6800	6800	6800	6800	6800	6800
6900 lbs.	6900	6900	6900	6900	6900	6900	6900	6900	6900
7000 lbs.	7000	7000	7000	7000	7000	7000	7000	7000	7000
7100 lbs.	7100	7100	7100	7100	7100	7100	7100	7100	7100
7200 lbs.	7200	7200	7200	7200	7200	7200	7200	7200	7200
7300 lbs.	7300	7300	7300	7300	7300	7300	7300	7300	7300
7400 lbs.	7400	7400	7400	7400	7400	7400	7400	7400	7400
7500 lbs.	7500	7500	7500	7500	7500	7500	7500	7500	7500
7600 lbs.	7600	7600	7600	7600	7600	7600	7600	7600	7600
7700 lbs.	7700	7700	7700	7700	7700	7700	7700	7700	7700
7800 lbs.	7800	7800	7800	7800	7800	7800	7800	7800	7800
7900 lbs.	7900	7900	7900	7900	7900	7900	7900	7900	7900
8000 lbs.	8000	8000	8000	8000	8000	8000	8000	8000	8000
8100 lbs.	8100	8100	8100	8100	8100	8100	8100	8100	8100
8200 lbs.	8200	8200	8200	8200	8200	8200	8200	8200	8200
8300 lbs.	8300	8300	8300	8300	8300	8300	8300	8300	8300
8400 lbs.	8400	8400	8400	8400	8400	8400	8400	8400	8400
8500 lbs.	8500	8500	8500	8500	8500	8500	8500	8500	8500
8600 lbs.	8600	8600	8600	8600	8600	8600	8600	8600	8600
8700 lbs.	8700	8700	8700	8700	8700	8700	8700	8700	8700
8800 lbs.	8800	8800	8800	8800	8800	8800	8800	8800	8800
8900 lbs.	8900	8900	8900	8900	8900	8900	8900	8900	8900
9000 lbs.	9000	9000	9000	9000	9000	9000	9000	9000	9000
9100 lbs.	9100	9100	9100	9100	9100	9100	9100	9100	9100
9200 lbs.	9200	9200	9200	9200	9200	9200	9200	9200	9200
9300 lbs.	9300	9300	9300	9300	9300	9300	9300	9300	9300
9400 lbs.	9400	9400	9400	9400	9400	9400	9400	9400	9400
9500 lbs.	9500	9500	9500	9500	9500	9500	9500	9500	9500
9600 lbs.	9600	9600	9600	9600	9600	9600	9600	9600	9600
9700 lbs.	9700	9700	9700	9700	9700	9700	9700	9700	9700
9800 lbs.	9800	9800	9800	9800	9800	9800	9800	9800	9800
9900 lbs.	9900	9900	9900	9900	9900	9900	9900	9900	9900
10000 lbs.	10000	10000	10000	10000	10000	10000	10000	10000	10000

develop a panel station 1. We prefer to handle them in another way however. For the present their location will be modified a panel point and they will be called a and b as in case in the table. Later in the individual analysis dotted line will be drawn from panel points a and b to D and E, and as shown due to these surrounding weights moved along the dotted lines to the fuselage structure proper. This will give a lot of arithmetic and will give the true loads due to their weight a and b.

The oil tank lies between four panel points. This is a common case in many airplanes that use the Pratt type for the engine. In this case the load is first distributed horizontally between stations 1 and 2. The load is distributed in proportion to its distance from the stations. Then the load should be distributed vertically between the upper and lower panel points at each station. It so happens that panel point 11, a, is on the same level as the oil tank and therefore the whole load of 55 lb. that reaches station 1 stays right there. The remainder of the load of 26.6 lb. which reaches station 2 is broken up and distributed inversely as its distance from panel points 11 and 12.

It is particularly important, as mentioned before, to keep the moments of the panel loads the same as those of the original weights. This applies especially to the horizontal moments but it is not so necessary to be so careful of the vertical moments. In some cases, as wings, gun and tanks, chassis, tail skid, and tail group which come just above or below the fuselage, we may consider them as the load. Then they are merely distributed between the panel points in either side inversely as their distance away. The error we are making here in the vertical moments is not as great as it seems as moving the chassis up to the lower longerons compensates for moving the loads above the upper longerons, due to this level. Our center of gravity as a result moves very little if at all. It is to be noted how easily the tail group was placed at a panel point. If we had placed the center of gravity behind the tail post we would have had to add a panel point a and handled it in the same manner as panel points a and b for the propeller and engine. If we had shown the center of gravity forward of the tail post the tail weight of 50 lb. could have been brought down to the upper longerons and distributed between panel points 12 and 13.

The distribution of the fuselage weight itself among panel points is probably the most difficult to handle. The only method available for doing this is to arbitrarily break the fuselage weight up into small weights and apply one at each panel point putting the larger weights in the region of the skin. The structure in this region will be much heavier than in the rear and usually the fuselage weight is broken into small weights which are located in the cabin. After distributing these loads at the panel points their moment must be taken about the datum line. The sum of all these moments must vary exactly equal the total fuselage moment. Verify this will put the case on the first station as the first load must be shifted slightly. Obviously in making out the panel loads the fuselage panel loads due to the fuselage weight itself were all five pounds from station 9 back. This was a moment of 21,263 lb. ft. for all the panel loads as against an original fuselage moment of 15,120 lb. ft. This is too great a difference. By moving one panel from panel point 12 to panel point 35 the moment of the panel loads became 15,120 lb. ft. which is very close to the original moment. This distribution is therefore satisfactory.

The error which we have not as yet accounted for is located between these panel points. These distribution among the surrounding panel points must be handled quite differently. The method we have taken the passenger weight of 330 lb. is shown in the figure by point P. Panel P is the figure



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is loaded accurately with respect to the surrounding panel loads. Fig. 16 is merely one little section taken from Fig. 15 and represented very schematically. Perpendicular lines are dropped from Point P to each of the three sides and from each panel point to the opposite side. To obtain the



Triangular Distribution
of
Panel Loads.

Fig. 16.

proportion of the 330 lb. on each panel point the following method is used:

Panel Point (5):	$330 \times \frac{50/254}{80/254} = 116 \text{ lb.}$
Panel Point (6):	$330 \times \frac{66/254}{80/254} = 110 \text{ lb.}$
Panel Point (7):	$330 \times \frac{48/250}{80/250} = 102 \text{ lb.}$
Total:	330 lb.

If a check is made it will be found that each of these panel loads times its area when summed up will equal the original moment of the 330 lb. weight times its arm of 162 in. It is to be noted in computing any particular panel load we use the line dropped from it to the opposite side and its line from P parallel to that line. The power plant engine, pilot, and baggage distribution had to be figured in the same way for the panel load table. With a Warren truss fuselage structure this method will always have to be used for load falling within the triangles. The measurements marked on the figures are as accurate as the figures were originally drawn. It is likely that it will be reduced in publishing but all its dimensions will bear the same relation they now do.

Summation of Panel Loads.

When the table of panel loads has been completed a schedule is told them as has been done at the bottom of the table. This will locate any stress that may have been made. The horizontal moment of each panel load must also be found and the sum of all of them found. If this is fairly close to the original moment we will not have much trouble in making our stress polygon close. If the difference is the two moments is more than 200 in. lb. it is very likely that a mistake has been made somewhere and the work should be done back. It is worth while spending a great deal of time getting the panel loads distributed correctly as it greatly facilitates later work.

It is to be remembered that these panel loads are for a unit weight and do not include any load factor as yet. In the next chapter the factor remaining under conditions will be analyzed and the fuselage members designed.

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To be continued in the next issue of AVIATION

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N.A.C.A. Progress

Continued from page 2523

as to what form of cooling and fuselage shape allow of the best cooling of the engine.

The point is of the open throat type, but the stream of air flowing through the room in which the cylinders stand is very sharply deflected, so that the air on the floor is practically stationary, while about within reach of one's hand it is blowing at over 120 m.p.h. When the tunnel is in operation and the plane's engine is running, the noise in pulsing and eddies are given by a system of signals. When the tunnel was first built, not enough allowance had been made for the low pressure in the tunnel room, and the walls nearly caved in, but since then they have been reinforced.

Pressure Run Up to 200 Lb. Per Sq. In.

In the aerodynamic engineer, the variable pressure wind tunnel is one of the most interesting things at Langley. To handle where ordinary atmospheric pressure is used, a substitute must be made for the model effect. In other words, the full sized airplane does not have a performance which can be directly calculated from data gathered from the small model, but certain corrections must be made, and these are in a direct ratio to a number of elements. By increasing the



The 100 Model propeller mounted in the N.A.C.A. 20 ft. wind tunnel.

pressure of the air and therefore its density, it is thought that small scale models will give the same results as those obtained in full sized airplanes. So far there is reason to believe that the variable density tunnel will enable engineers to obtain actual performance from small models considerably more accurately than can be done at present. Pressure



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in the tunnel ran up to 200 lb. per sq. in. and the turbulent diffusion of designing and operating the tunnel was very considerable. Last year the tunnel caught fire and the inside was completely burned out before the pressure could be released. In fact, the heat caused the pressure to be increased and a real disaster was narrowly averted.

In the same building as the variable density tunnel are two entrances to the study of objects in a moving fluid. One is a water tank with glass sides. The water is circulated comparatively slowly and coloring matter is introduced. This enables one to see the flow about any object which is introduced. As the density of the water is very much greater than that of air, it can be circulated very slowly and yet produce the same phenomena of turbulence, etc., as are produced by a very high speed air flow. This method of testing is inexpensive and enables the student to visualize the disturbances caused by a disturbance. The other apparatus is a jet of air which is forced through a small tunnel at velocities above those of sound. At present, only propeller tips even approach such velocities, but there is an underground apparatus field here which has interesting possibilities.

Experimenting With a Variable Stroke Engine

The engine division of the N.A.C.A. is conducting a series of experiments with a variable stroke engine. This enables the compression to be changed while the engine is in operation, and much can be found out about the anti-distorting qualities of various fuels. Work is also being done on heavy air engines. In order to do this, a new engine design has been developed to photograph the diffusion of the fuel air which is injected into the cylinders.

During the past year, tests have been run to determine the pressure on the bottom of airplane flaps on landing. These

show that the maximum pressures occur at the step when seven pounds to the square inch were registered. There was no pressure leakage and the accelerometer showed that the impact was about six times that of the weight of the plane. The full scale experiments on increasing the landing angle by sucking or pumping air in the area behind the landing flap on the upper surface of the wing has not been pushed forward vigorously, as the results obtained were not at all promising.

Some interesting data on the formation of ice on planes has been gathered and many flights were made through clouds, where ice forming conditions occurred. It is hoped to build a small tunnel which can be refrigerated for further study of this subject.

One of the most interesting pieces of work which the flight testing section has been doing is the recording of the pressure on the wings and control surfaces of a parrot plane during violent maneuvers. There are now in all parts of the plane and sections of pressure are transmitted to recording instruments from which pressure distribution charts are made. This work is of course of enormous value from the structural standpoint in the design of present planes. Certain parts of the wing showed lift of 200 lb. per square foot. The most spectacular work of the section was the tests conducted by Luke Christopher to find the maximum strains which could be placed on a plane. A Curtiss Pursuit pulled out sharply from a dive at 150 m.p.h. showed stresses of one-eighth times those found in normal flight. At the time of pulling out of the dive, the forces acting on the pilot were as great as to force him backwards in the eyes and serious internal hemorrhages.

At present the flight section is testing a Consolidated transport plane which is fitted with Handley Page slots. Many of

the existing experiments first in the plane and were greatly improved. Comments were varied, but it was generally concluded that the plane was more stable beyond the stalling angle and somewhat more controllable. During the tests Handley Page slots went up to the Handley Page slots of the Consolidated. Both planes were stalled and, as far as an observer on the ground could see, they both acted as if about the same speed and both kept on an even keel. The two planes are too different to supply a real comparison, and it is concluded the Handley Page has several lateral control

What Tamed Studies of Airplanes

Springing of safety devices, the N.A.C.A. has also been making some very interesting wind tunnel studies of airplanes of several types, of the Triplane type, and of various combinations of wing and tail. They have also developed a wing section with an extremely flat top to the lift curve. From about 100 to over twenty degrees of incidence the lift stays about the same. This means that in flight a plane would not stall suddenly, and also when landing it would have much less tendency to "float" for long distances, because it would slide back before the stallable stalling point.

The above are only a few of the more important points of the work being done by the N.A.C.A. The field of their activity is very wide, but the dissemination of it to the aviation world was very limited. At the beginning of the conference, each department had outlined briefly what he was doing. Then the visitors were split up into parties and visited the various laboratories. Everything was well labeled and charts explained what had been accomplished. After everything had been seen and explained, a general discussion was held and suggestions were registered as to what development work should be undertaken during the course of the next year. Secretary Warner closed the keynote when he said that the efforts of the N.A.C.A. should be turned towards solving those problems which are retarding the commercial progress of aviation. Research, he said, could properly be done in showing planes in improving ventilation in cabin planes, and in airplane more comfortable chairs.

The meeting was not only instructive but was also a very pleasant and well handled occasion and many felt that it could well have been extended over another day.

Rules of the 1928 National Air Tour

Continued from page 1232

apply for the difference in gasoline and oil consumed during the flight.

(d)—No contestant will be allowed to carry passengers in excess of the original designed seating arrangements of the plane as reflected in its sections 24-31 and 50-51 of the Air Commerce Regulations.

(10) (a)—The Contest Committee will require each plane to qualify by flying back bases over a measured course of approximately one mile, twice in each direction to establish maximum speed to be used in the future.

(b)—These speed trials will be held, at the option of the Contest, at any time during, before or after the stock and stock trials. After the beginning of any of the tests no change will be permitted to the propeller setting of a plane, and all of its parts must be completed.

(c)—In order to discourage racing and its attendant dangers, the maximum speed will be used in the future, unless the average speed for an individual lap shall fall below 50 per cent of the maximum credited speed. In such an event



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Col. Harold E. Hartney, American "Ace" says: "I most cordially urge you to take Home Study in Aviation. It gives an excellent basis for anyone contemplating either the aviation of aviators or military service in the field."

Ray J. Lewis, Aviation Editor, Cleveland, writes: "This Course gives to the layman, in simple language, answers to questions which exist in the minds of beginners."

La. Leigh Wade, Read the World-Flair, says: "This Course is a wonderful exposition of the fundamentals of aviation. I can readily recommend it to those seeking authentic information."

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(21) Fifteen minutes before the actual starting time the official starter shall raise a red flag which shall be used as an alarm to start the engine and, where permitted, to see their places to the proper position for the official start. Tearing about the field other than for the purpose of getting into proper position is the starting line shall not be permitted.

(22) One minute before the starting time of the first pilot, the official starter shall raise a white flag which shall be dropped at the proper time to indicate go-ahead of the first pilot. Other contestants will be given a starting signal at various intervals thereafter and this shall be designated to suit one by the dropping of the white flag.

(23) Should a contestant be unable to start at the time the signal is given he shall be permitted to depart at any time thereafter but the departing time shall be figured from the time of the signal and not from the time of the starting signal. In this event he shall not interfere with the starting of any other pilot but shall await the starter's signal that he can start any time. This signal may be given by the starter at any favorable time and need not be delayed until after the departure of the last contestant remaining.

(24) Entries will at all times conform to the Air Traffic rules set down by the Director of Aeronautics, Department of Commerce.

(25) Each pilot arriving at a second station shall be shot for other planes which may be in the act of landing or crossing and upon landing each pilot shall taxi from the landing area immediately and proceed to his proper position in the line as indicated by the director.

(26) Each contestant shall be seated on a bench of elapsed time which shall be taken from the starting signal at the previous control and completed when the contestant shall cross the finish line at the next control in full flight. The actual landing time at each control shall not be recorded as it may be determined by each pilot for himself and need not be used as to present contestants on a result thereof.

(27) Each pilot will carry an official log which will be provided by the Tour officials. Prior to the start from Dearborn, this log must be filled out and contain all information regarding the entry together with a record of passengers not weight of fuel and baggage carried. At each control of the tour, this log must be turned over to the clerk who will enter thereon the record of his checking and note thereon any discrepancy he may discover. Immediately thereafter the pilot shall turn the log over to the official scorer who will retain it until after the day's score is completed. If such time be returned to the pilot who must present it to his club before departure on the following day. The starter will then advise O.R. the log or make a notation of any discrepancy he may have discovered at this time and it shall be the duty of the pilot to see that this is done. In the event of failure on the part of the pilot to obtain an official O.R. or in the event the pilot will at the next control, at the discretion of the referee, be penalized not less than fifty points but in no case in excess of the total number of points for the log in which such penalty occurred.

Deferred Start.

(28) The start of the first and subsequent planes from each of the several controls may be deferred only at the discretion of the referee and only in the event of technical trouble or other questionable conditions. If it appears that there is any reason for delaying the start, the referee shall consult with the pilot and if no protest is recorded, the start shall not be deferred. The starting time for the first plane from each control shall be in accord with the 1925 Tour Schedule it laid down by the officials in charge.

Eligibility of Pilots.

(29) Any designated pilot shall be eligible to compete in

the 1928 National Air Tour, provided he can show evidence of having complied with the requirements of the Aeronautics Division of the United States Department of Commerce and still in addition thereto, hold an F.A.I. certificate and the second sporting license issued by the Contest Committee of the National Aeronautic Association. Any pilot may, however, be relieved from duty or suspended by an official representative of the United States Department of Commerce or by the referee for physical inability. Such suspension by the referee, however, shall not be made without consultation with a reputable physician and upon the advice of such physician.

(30) Any pilot may be required to undergo a physical examination within 48 hr. prior to the start of the Contest or at any time thereafter at the discretion of the referee. No pilot shall be permitted to compete who is not in the opinion of the examining physician, in physical condition to do so without injury to himself or other contestants. Contestants shall conduct themselves in an orderly and gentlemanly manner throughout the entire Tour and for failure to conform to the general rules of the Contest or for conduct detrimental to the Tour or to successful aviation generally, may be, in the discretion of the referee, fined in a sum not to exceed \$5 for each infraction. Since the entrant is responsible for the action of the contestants, this fine may be deducted from any money or guarantee money at the discretion of the Tour.

(31) Rules of conduct applying to contestants shall also apply to officials and ground crew. Any official conducting himself in a manner which, in the opinion of the officials at a nearby station, shall be detrimental to the Tour, shall be dropped therefrom and the contestant carrying such official or one shall not proceed further with his but shall replace such official with another passenger or with ballast as required.

Preparations and Tests Prior to Start

(32) All entries, consisting of planes and pilots, shall be required to be at the Ford Airport at Dearborn, Michigan, not later than 8 A.M., on Wednesday, June 27. This is to permit clerks to weigh up planes and complete the work of loading and sealing tanks, fuel and spare parts in planes in preparation for the stock and stretch tests which will be held on Thursday, June 28. No flying will be permitted on June 27 which will conflict with the weighing or loading of planes.

(33) Beginning at 8 A.M. on Thursday, June 28, the stock and stretch tests will be held. At this hour, each plane will be up in the order of entry, equipped with full control load during the stock and stretch tests and the maximum speed test, at the discretion of the referee. At each control, all oil tanks must be filled supplying for reserve air space which may not exceed 25 per cent. of the total capacity of the tanks. In addition to fuel and oil, each plane in the stock, stretch and maximum speed tests must carry full useful load permitted by the Department of Commerce of the United States as specified in the approved type certificate awarded the plane. In addition to the gasoline and oil, this load may, for the purposes of these tests, be in the form of passengers or ballast in the pilot bay alone.

After the stock, stretch and maximum speed tests have been completed, a contestant will be allowed to reduce his useful load to 75 per cent. of the maximum permitted by the Department of Commerce and specified in the approved type certificate awarded the plane. No contestant shall be allowed to carry gasoline except in tanks regularly provided for such purpose and which shall have been used for such purpose during the stock, stretch and maximum speed tests.

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particulars—fuelage consumption and loading; fuelage assembly, sheet metal work, and final assembly.

The Lincolnton Company is the oldest commercial airplane manufacturing company in the United States. It has gone through the trials and vicissitudes of pioneering aviation. Those troubles are in the past, now, and the future looks rosy. Under present progress and with the facilities offered by the more ample quarters, the 1939 production is slated to be 12 planes a week.

The Stinson "Junior"

Continued from page 1517

push rods work in graphite bearings mounted behind the rear spar.

The Warner Scarb engine, developing 116 hp. at 1900 r.p.m., is mounted in a welded steel tubular frame behind the fuelage frame. The engine is fitted with a Hartzel wooden propeller. Standard equipment includes exhaust regulator, and carburetor heater. The normal fuel capacity is 55 gal., which should permit a range of approximately 600 mi. Gasoline is carried in two wing tanks, mounted between the wing spars in the inner drag on each side of the fuselage. The tanks are supported by steel straps bolted to the spar and partly by the drag wire bracing. The supports are covered with felt to prevent rubbing against the tank. In addition to the wing tanks, there is a one gallon auxiliary tank mounted on the firewall. Two fuel lines from the wing tanks are connected to the top of this auxiliary tank and two fuel lines run down the rear and are connected to the bottom of the auxiliary tank. In this manner air is gas is eliminated and the one gallon tank will care for most of the engine starting any morning. A strainer is employed between the auxiliary tank and the carburetor. The oil tank is under the engine, working in front of the wind shield. It is tilted from the main on the right side.

The "J12" was designed according to the Department of Commerce specifications, with a factor of 0.95 for high incidence and 0.75 for low incidence. The landing chassis and tail wheel were designed for a load of 400 lbs.

The manufacturer's specifications are as follows:

Length overall	25 ft. 0 in.
Wing span	27 ft. 0 in.
Wing area	120 sq. ft.
Horizontal stabilizer area	20 sq. ft.
Vertical stabilizer area	12 sq. ft.
Propeller area	12 sq. ft.
Propeller pitch	12 in.
Propeller speed	1200 r.p.m.
Propeller diameter	48 in.
Propeller weight	120 lb.
Propeller moment	1200 in.-lb.
Propeller balance	1200 in.-lb.
Propeller vibration	1200 in.-lb.
Propeller noise	1200 in.-lb.
Propeller life	1200 in.-lb.
Propeller cost	1200 in.-lb.
Propeller maintenance	1200 in.-lb.
Propeller inspection	1200 in.-lb.
Propeller repair	1200 in.-lb.
Propeller replacement	1200 in.-lb.
Propeller disposal	1200 in.-lb.
Propeller storage	1200 in.-lb.
Propeller handling	1200 in.-lb.
Propeller transport	1200 in.-lb.
Propeller installation	1200 in.-lb.
Propeller removal	1200 in.-lb.
Propeller testing	1200 in.-lb.
Propeller certification	1200 in.-lb.
Propeller approval	1200 in.-lb.
Propeller acceptance	1200 in.-lb.
Propeller delivery	1200 in.-lb.
Propeller receipt	1200 in.-lb.
Propeller inspection	1200 in.-lb.
Propeller maintenance	1200 in.-lb.
Propeller repair	1200 in.-lb.
Propeller replacement	1200 in.-lb.
Propeller disposal	1200 in.-lb.
Propeller storage	1200 in.-lb.
Propeller handling	1200 in.-lb.
Propeller transport	1200 in.-lb.
Propeller installation	1200 in.-lb.
Propeller removal	1200 in.-lb.
Propeller testing	1200 in.-lb.
Propeller certification	1200 in.-lb.
Propeller approval	1200 in.-lb.
Propeller acceptance	1200 in.-lb.

SIDE SLIPS

By ROBERT R. OSBORN

A news item concerning the flight of Army aviators planes to the rescue of the stranded Bremen, stated that the rescue pilot might have to be dropped with a parachute, and that after inspection from the air "General Fochet will decide whether landing on and make a landing on area, an amphibious plane too dangerous."

If the situation looks doubtful, the safest procedure would seem to be an eight point landing on the wheels, tail skid, a pair of skis, the parashoot and wing tip floats.

Mr. H. W. W. of Winthrop, Mass., made long a clipping from a prominent Boston paper of an article about a passenger of various accidents. According to the item, one lady pilot has had "three French flights without the slightest mishap, to her credit."

Of course the expression "foiled flight" does sound funny at first, but when you stop to analyze it a large proportion of the flights are foiled. You know that feeling, "Both I don't feel like flying today but I don't feel much like starting either."

Captain Wilkins and Carl Carlson may not have discovered any land or navigational routes on their recent flight across the Arctic region, but they have at least discovered a possible due to the whereabouts of the mysterious Soviet plane. According to a Minneapolis clipping from H. F. C., their

great flight was from Point Barrow, Alaska, to Green Harbor, Switzerland.

For reasons which may be obvious upon reading it, the editor has transferred the following list of news from a syndicate correspondent to this column, from the second place under "Foreign News."

Thursday of last week the town of Frouin was attacked by the rear of the engine of a monoplane. The whole airplane plowed the air and with gaily rapidly cleared the town of any danger at this central. Moments later it landed in the grounds of the hotel. Upon their descent the refugees were given an immense ovation by the people gathered there.

Right away the aviator "Videman" R. B. Perry on his elegant machine conducted them to the telephone office where they met various dispatches. Later the airplane started, was lodged in the Hotel Frouin. By Grand Perry I was permitted to them and in a rapid interview I could note the following: Captain C. B. D. Calver is a special pilot of the Pan-American Airways who bought the plane from Henson and then handed it over to Victor Dufin, special pilot to fly the day while the work here. R. A. Smith, pilot and photographer who takes the photographs and who is chief of the flying squadrons of the Fairchild Aerial Survey. The machine is a monoplane, number 3690 constructed by the Fairchild Aircraft Co. of a special type to take pictures in the air. They are now making an aerial survey for the Cuba Co. and the United Fruit Co.

The aviators are all famous, having experience since the great world war and are all members of the Association of Quiet Birds, an international social club which has in members the most famous aviators who have arrived in the United States at Longbeary, Chamberlain, Lohrstein, Bird, Austin and others.

FOREIGN NEWS

By special arrangement with the Transportation Division Bureau of Foreign and Domestic Commerce

A.A.P. on Annual Distance Flights

The British Royal Air Force is completing its second long distance flight from Cairo to Cape Town and back. The plane used was four Napier III P lightness powered with Napier light engines. These planes are a military observation type fitted for day bombing, lighting, reconnaissance, photography, etc. It is designed to use either wheel or tail propellers.



The four Napier III P lightness used by the British Royal Air Force. The craft is powered with a Napier light engine.

too underpowered. Although this is a comparatively new type for the British Royal Air Force, many squadrons are now using them. It is stated that they have a high performance and are very maneuverable.

Poland Has New Air Traffic Rules

By treaty governments ruling, new air traffic regulations become effective this month in Poland. By the provisions of the decree aviation over Polish territory is permitted to flying machines registered in Poland and to Polish military planes and aircraft. Foreign-owned planes and aircraft are permitted to operate over Polish territory on the basis of international agreements or under special permit from the Polish authorities.

The Ministry of Communications is charged with the execution of private and commercial aviation, and the Ministry of War with military aviation. Pilots and crews of all Polish flying machines must be of Polish nationality and be at least 21 years of age at the time of pilot's license and every flying plane or air in the case of pilot's license.

Landing fields may be established and maintained only under permission from the Ministry of Communications, and flying machines must be accepted by the designated authorities before each flight. Ownership of foreign flying machines, and all legal acts performed thereon, are subject to regulation by the law of the country in which they are registered.

This decree, furthermore, includes regulations pertaining to passport and customs formalities.

Long French Flights Need Government O.K.

Airplane flights over great distances cannot be attempted from France to the United States without careful investigation and endorsement by the government, according to a recent decision of the Council of Ministers of France. The new ruling, however, is not expected to affect pilots who were preparing for trans-Atlantic flights as the new decree was made.

Dyke's Aircraft Engine Instructor

A N invaluable handbook on the construction, maintenance and repair of aircraft engines, accessories and instruments. The Wright Whirlwind, Pratt & Whitney Wasp, Curtiss engines, Packard engines and Fairchild-Camden Cars are described in detail with additional chapters on other modern engines and on war surplus engines. Accessories such as carburetors and carburetor jets, governors, Haywood injection starters, fuel batteries and various instruments are discussed in detail. There are also chapters on lubrication, maintenance and Department of Commerce regulations.

The author has utilized many books on aviation subjects and is a thoroughly trained authority on engine mechanics. All of the subjects are treated in the same popular fashion which makes reading and understanding the greatest source of information for the busy aviator. The World War. Price, \$2.50.

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John Leonard, C. A. Tinsler, Mark Alexander, M. H. Cullman, and Mark D. Moore. This group is also trying to get their city designated as a direct point of exchange on the proposed air line to connect the Twin Cities and St. Louis, Ill.

Detroit, Mich.

By John T. Neill

More than 325 airmen were invited to enter a prize design contest held by the Detroit Board of Commerce aircraft events committee by way of selecting an airplane design for the Detroit Air-Olympics, a name recently chosen for the National Air Tour, the Gordon Bennett International Balloon Race, and the Boys National Model Airplane Contest, all of which are to be held here June 30.

The contest closed May 23, with Ray Cooper, manager of aircraft events for the board, Earl G. Wells, assistant manager, and John S. Hildreth, chairman of the contest committee, as judges. A prize of \$300 was given the winner.

George Haldeman, president of the board, now with the St. Louis Aircraft Corp., and William A. Moore, secretary of the corporation, were guest speakers before the Post House Lions Club. Many recently returned from Syracuse, when the Syracuse company exhibited in the New York state armory show.

Stability as applied to aircraft was discussed recently by Prof. A. Mueller, assistant in aeronautics at the University of Michigan, who spoke before the Detroit Flying Club.

The Royal Oak Flying Club, whose roster includes 41 members, has rented 72 acres of land near Detroit to be used as the club's landing field. Scott Timmer Carroll of the Royal Oak Police Department and member of the club, recently let the club's first plane, a Waco 10, from Tynanville, where it was purchased from the Kasper Flying Service, Waco distributor for Michigan.

Chicago, Ill.

By N. H. Shuman

To enable the owner of the *Pratt & Whitney* to enter in "best" more readily, that magazine has added a three-page complete equipment to its equipment. It will be used in the future, various pictures and meetings.

Money, also, "the flying business" has been engaged in a project. *Pratt & Whitney* was established in 1941 and is now in its 100th volume, having passed through four "transformations" of transportation. The first other than land-based. This one comes equipped, in its equipment, and was the airplane.

Appala Cammery, Knight Templar, of Chicago, plans to fly by airplane to the 1915 transatlantic contest of the St. Louis air line, which starts in Detroit in July. Officers, his companions and a hand are scheduled for the flight. Lloyd Bushy, president of the Chicago City Railway, recently took a flight by plane over the principal route of the city's street car system, the better to grasp the "lay of the land," as he said. Next time, he promised, he will take along some maps and a theodolite as well, in order to make a permanent record of the data available on such a trip.

Toussaint, N. D.

John's Airport here is among some activity with the spirit more visible. The Leach Brothers are including an all-Japan, adding a pair of high lift wings they have purchased. Bill Leach has made a complete steel landing gear for the plane.

Jerry France recently dropped in with his Standard (pilot) by LaPorte of Mackinac. The two are considering establishing a flying school at Tynanville.

John Bertram has commenced to build up his Home Field and, Bertram, who maintains the airport, extends invitation to all pilots who come to the vicinity to pay the field a visit. Two hangars are at the airport, and the hangars are open.

May 23, 1935

Waco, N. D.

By John T. Neill

Waco Airport here is showing spring activity with four planes just being in a recent landing in passenger carrying. The planes are a Ryan monoplane, owned by the Waco and Oil Co. of Waco and piloted by Carl Skoug, a Waco 10 owned by the Dakota Airways Co. and piloted by Carl Skoug, a Curtiss with floats at the controls, and Tynanville's C-1A biplane piloted by LaPorte.

The Dakota Airways Co. has opened a school and reports a number of students. Most will be a Ford Tour stop this year, and preparations for the event are already under way.

Manfield, O.

Manfield's new municipal airport was formally dedicated in a recent celebration. The event was made a half-day affair at the city, business closed and schools letting out for the occasion.

Speeches were in attendance at the dedication. The list of those in attendance included, Cleveland, Dayton, and Akron. The Goodrich company's hangar "Plymouth" also arrived at the field. Clarence Chamberlain, New York to Cleveland, was the honored guest.

A special air show and demonstration was put on by the firm to give the afternoon as a part of the ceremony.

Superior, Wis.

The Superior Field, formerly known as the Arrowhead Airport, has been in poor condition due to the late spring, but now guaranteed runways will be provided, take off and landing facilities will be mostly completed. An OX-51 Tri-Plane at the Both-Thompson Airway of St. Paul and a Fordville cabin, monoplane owned by the Mid-West Sales & Trade Co. of Minneapolis made a trip to the city with passengers.

C. M. Peterson of Minneapolis captured Ken Pratt, who is now with W. E. (Doc) Stohrer as chief pilot, as the pilot for the Superior Airways. Student instruction in ground, passenger, baggage and emergency trips are made.

Waukegan, Wis.

By F. E. Turner

All a recent meeting of the executive committee of Northern Wisconsin Airport, Superior, reports lighting facilities were ordered by Alexander Airport here. These include a 1,000,000-watt power house. With the new equipment installed, flights will be turned on for visiting pilots who certify the airport is adequate for their coming. Assessment is also made that a new hangar has been planned in the hangar and for new room and hangar room facilities are being completed.

Vancouver, Wash.

Municipal, county, or port ownership of the Vancouver, Wash., airport has been proposed by the local chamber of commerce. The field is now under lease to the chamber from the Spokane, Portland & Seattle Railroad. The field is generally used by commercial fliers who are not allowed to operate from the Pearson Field, the Army airport.

Capt. J. E. Eagle has been assigned to Pearson Field here in connection with the Navy. Captain Eagle came here from the Field, San Francisco.

Elvert, Wash.

The community under a hangar for some time tonight at night in Seattle, has given 20 to be a highly awarded city, according to "The" Rankin, who recently visited the Elvert Airport. Elvert's field is up-to-date, with \$10,000 invested in hangars and equipment. Forty-two students are enrolled in a school taught by John Scott.

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Treats on aerodynamics and its application to airplane design, divided into three parts: static and aerial characteristics, performance characteristics and stability (1937) (revised edition).

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